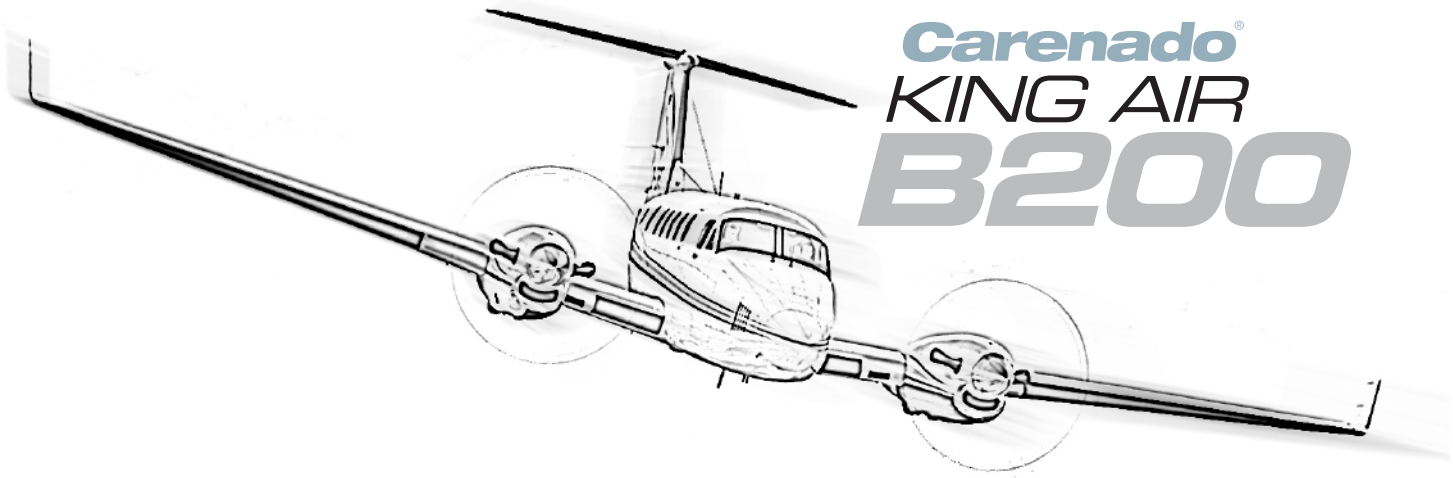
Performance Tables



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NORMAL CRUISE POWER

1700 RPM

ISA – 30 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	-10	-15	2230	100	930	240	228	241	229	242	230	242	231
2000	-14	-19	2230	100	906	237	232	238	233	239	234	240	235
4000	-18	-23	2230	100	882	235	236	236	237	237	238	238	239
6000	-22	-27	2230	100	858	233	241	234	242	235	243	236	244
8000	-25	-31	2230	100	834	231	245	232	246	233	248	234	249
10,000	-29	-35	2230	100	814	228	250	230	251	231	252	231	235
12,000	-33	-39	2230	100	794	226	255	227	256	228	257	229	258
14,000	-36	-43	2230	100	780	224	259	225	261	226	262	227	263
16,000	-40	-47	2230	100	768	221	265	222	266	224	268	225	269
18,000	-44	-51	2230	100	760	219	270	220	271	221	273	222	274
20,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
22,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
24,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
26,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
28,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
29,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1700 RPM

ISA – 20 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	0	-5	2230	100	934	238	231	239	232	240	232	241	233
2000	-4	-9	2230	100	908	236	235	237	236	238	237	239	238
4000	-8	-13	2230	100	880	233	239	234	240	235	241	236	242
6000	-11	-17	2230	100	858	231	244	232	245	233	246	234	247
8000	-15	-21	2230	100	838	229	248	230	250	231	251	232	252
10,000	-19	-25	2230	100	818	226	253	228	254	229	255	230	257
12,000	-23	-29	2230	100	796	224	258	225	259	226	260	227	262
14,000	-26	-33	2230	100	782	222	263	223	264	224	266	225	267
16,000	-30	-37	2230	100	772	219	268	221	270	222	271	223	272
18,000	-34	-41	2230	100	764	216	273	218	275	219	276	220	278
20,000	-37	-45	2230	100	758	213	278	215	280	216	281	218	283
22,000	-41	-49	2141	96	724	207	278	209	281	211	283	212	285
24,000	-45	-53	2007	90	680	199	276	201	279	203	282	205	284
26,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
28,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
29,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1700 RPM

ISA – 10 ℃

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	10	5	2230	100	940	236	233	237	234	238	235	239	236
2000	6	1	2230	100	914	234	238	235	239	236	240	237	240
4000	3	-3	2230	100	888	232	242	233	243	234	244	235	245
6000	-1	-7	2230	100	862	229	247	231	248	232	249	232	250
8000	-5	-11	2230	100	840	227	251	228	253	229	254	230	255
10,000	-9	-15	2230	100	818	225	256	226	257	227	259	228	260
12,000	-12	-19	2230	100	800	222	261	223	262	225	264	226	265
14,000	-16	-23	2230	100	786	220	266	221	268	222	269	223	270
16,000	-20	-27	2230	100	776	217	271	218	273	220	274	222	276
18,000	-24	-31	2230	100	768	214	276	216	278	217	279	218	281
20,000	-27	-35	2141	96	734	208	276	210	279	211	281	213	283
22,000	-31	-39	2052	92	698	201	276	203	279	205	281	207	284
24,000	-35	-43	1918	86	652	193	274	195	277	197	280	199	283
26,000	-40	-47	1784	80	606	184	270	187	274	189	278	191	281
28,000	-44	-51	1650	74	558	174	265	177	270	180	275	183	278
29,000	-46	-52	1583	71	534	168	261	172	267	176	272	179	277
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1700 RPM

ISA

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	20	15	2230	100	946	235	236	236	237	237	238	238	239
2000	16	11	2230	100	920	232	240	234	241	235	242	235	243
4000	13	7	2230	100	892	230	245	231	246	232	247	233	248
6000	9	3	2230	100	866	228	249	229	251	230	252	231	253
8000	5	-1	2230	100	844	225	254	226	255	228	257	228	258
10,000	1	-5	2230	100	822	223	259	224	260	225	262	226	263
12,000	-2	-9	2230	100	800	220	264	222	266	223	267	224	286
14,000	-6	-13	2230	100	786	218	269	220	271	221	272	222	274
16,000	-10	-17	2230	100	776	215	274	216	276	218	277	219	279
18,000	-14	-21	2141	96	742	208	274	210	276	212	279	213	280
20,000	-18	-25	2052	92	706	202	274	204	277	206	279	207	281
22,000	-21	-29	1940	87	672	195	274	197	277	199	280	201	282
24,000	-26	-33	1829	82	628	187	271	189	275	192	278	194	281
26,000	-30	-37	1695	76	580	177	267	181	272	183	276	186	279
28,000	-34	-41	1561	70	534	167	261	171	267	175	272	177	276
29,000	-36	-42	1494	67	512	161	256	166	264	170	269	173	274
31,000	-40	-46	1360	61	472	149	246	156	256	160	264	164	270
33,000	-45	-50	1249	56	432	– – –	– – –	143	245	150	155	155	265
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1700 RPM

ISA + 10 °C

NOTE: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.

TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	30	25	2230	100	952	233	238	234	239	235	240	236	241
2000	27	21	2230	100	924	231	243	232	244	233	145	234	246
4000	23	17	2230	100	898	229	247	230	249	231	250	232	251
6000	19	13	2230	100	874	226	252	227	253	228	255	229	256
8000	15	9	2230	100	848	224	257	225	258	226	259	227	261
10,000	12	5	2230	100	826	221	262	222	263	224	265	225	266
12,000	8	1	2230	100	804	219	267	220	269	221	270	222	271
14,000	4	-3	2230	100	790	216	272	217	273	219	275	220	276
16,000	0	-7	2163	97	756	210	273	212	275	213	277	215	279
18,000	-4	-11	2052	92	714	203	273	205	275	207	277	208	279
20,000	-8	-15	1962	88	680	197	273	199	276	201	278	202	281
22,000	-12	-19	1851	83	648	189	271	192	275	194	278	196	281
24,000	-16	-23	1739	78	604	181	268	184	273	186	276	189	280
26,000	-20	-27	1606	72	558	171	263	175	269	178	273	180	277
28,000	-24	-31	1472	66	512	160	256	165	263	169	269	172	274
29,000	-26	-32	1405	63	492	154	251	160	259	164	266	167	271
31,000	-31	-36	1293	58	452	139	234	149	251	154	260	158	267
33,000	-36	-40	1160	52	414	– – –	– – –	133	234	143	251	149	261
35,000	-40	-44	1070	48	380	– – –	– – –	– – –	– – –	128	233	138	251

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1700 RPM

ISA + 20 ºC

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	40	35	2230	100	958	323	241	233	242	234	243	235	244
2000	37	31	2230	100	932	229	245	231	247	232	248	233	249
4000	33	27	2230	100	94	227	250	228	251	229	252	230	253
6000	29	23	2230	100	878	225	255	226	256	227	257	228	258
8000	25	19	2230	100	854	222	260	223	261	224	262	225	263
10,000	22	15	2230	100	832	220	265	221	266	222	268	223	269
12,000	18	11	2208	99	806	216	269	218	271	219	272	220	274
14,000	14	7	2219	95	762	210	269	212	271	213	273	215	274
16,000	10	3	2029	91	722	203	269	205	272	207	274	209	276
18,000	6	-1	1940	87	682	197	269	199	272	201	275	202	277
20,000	2	-5	1873	84	654	191	270	194	273	195	276	197	279
22,000	-2	-9	1874	80	624	184	270	187	274	189	277	919	280
24,000	-6	-13	1673	75	582	175	266	179	271	182	275	184	279
26,000	-10	-17	1539	69	536	165	260	170	266	173	271	176	275
28,000	-14	-21	1405	63	492	153	250	159	259	163	265	167	271
29,000	-17	-22	1338	60	472	145	242	153	254	158	262	162	268
31,000	-21	-26	1227	55	432	– – –	– – –	140	242	149	255	153	263
33,000	-25	-30	1115	50	400	– – –	– – –	– – –	– – –	136	243	143	258
35,000	-29	-34	1026	46	368	– – –	– – –	– – –	– – –	– – –	– – –	131	244

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1700 RPM

ISA + 30 ºC

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	51	45	2230	100	964	231	243	231	244	232	245	233	246
2000	47	41	2230	100	936	228	248	229	249	230	250	231	251
4000	43	37	2230	100	910	226	252	227	254	228	255	229	256
6000	39	33	2230	100	882	223	257	224	259	225	260	226	261
8000	36	29	2230	100	858	221	262	222	264	223	265	224	266
10,000	32	25	2163	97	818	216	264	217	266	219	268	220	269
12,000	28	21	2074	93	764	209	264	211	266	212	268	214	270
14,000	24	17	1985	89	724	203	265	205	267	206	269	208	271
16,000	20	13	1896	85	686	196	265	198	268	200	270	202	272
18,000	16	9	1806	81	650	190	265	192	268	194	271	197	273
20,000	12	5	1762	79	626	185	266	187	270	189	273	191	276
22,000	8	1	1717	77	602	179	267	182	271	184	275	186	278
24,000	4	-3	1583	71	560	169	262	173	268	176	272	179	276
26,000	0	-7	1472	66	516	159	255	164	262	168	268	171	273
28,000	-4	-11	1338	60	474	147	244	153	255	158	263	162	269
29,000	-7	-12	1293	58	454	137	232	147	250	153	259	157	266
31,000	-12	-16	1160	52	414	– – –	– – –	131	231	142	250	148	260
33,000	-16	-20	1048	47	382	– – –	– – –	– – –	– – –	126	230	137	251
35,000	-20	-24	959	43	350	– – –	– – –	– – –	– – –	– – –	– – –	121	230

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1700 RPM

ISA + 37 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	58	52	2230	100	968	229	245	231	246	232	247	232	248
2000	54	48	2230	100	940	227	249	228	251	229	252	230	253
4000	50	44	2230	100	912	225	254	226	255	227	257	228	258
6000	46	40	2230	100	886	222	259	223	260	224	262	225	263
8000	42	36	2141	96	838	216	260	218	262	219	263	220	265
10,000	39	32	2074	93	790	210	261	212	263	214	265	215	267
12,000	35	28	1962	88	738	204	261	205	263	207	265	209	267
14,000	31	24	1873	84	700	197	261	200	264	201	266	203	268
16,000	27	20	1806	81	664	191	261	194	265	196	267	198	270
18,000	23	16	1695	76	630	185	261	188	265	190	268	192	271
20,000	19	12	1695	76	606	180	263	183	267	185	270	187	273
22,000	15	8	1628	73	582	174	263	177	268	180	272	182	275
24,000	11	4	1539	69	542	165	258	169	264	172	270	175	274
26,000	7	0	1405	63	500	154	251	160	259	164	266	167	271
28,000	2	-4	1293	58	460	140	237	149	251	154	260	158	266
29,000	0	-5	1227	55	440	– – –	– – –	142	244	149	256	154	263
31,000	-4	-9	1137	51	406	– – –	– – –	– – –	– – –	137	245	144	257
33,000	-8	-13	1048	47	376	– – –	– – –	– – –	– – –	– – –	– – –	133	246
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA – 30 ºC

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	-10	-15	2230	100	930	240	228	241	229	242	230	242	231
2000	-14	-19	2230	100	906	237	232	238	233	239	234	240	235
4000	-18	-23	2230	100	882	235	236	236	237	237	238	238	239
6000	-22	-27	2230	100	858	233	241	234	242	235	243	236	244
8000	-25	-31	2230	100	834	231	245	232	246	233	248	234	249
10,000	-29	-35	2230	100	814	228	250	230	251	231	252	231	235
12,000	-33	-39	2230	100	794	226	255	227	256	228	257	229	258
14,000	-36	-43	2230	100	780	224	259	225	261	226	262	227	263
16,000	-40	-47	2230	100	768	221	265	222	266	224	268	225	269
18,000	-44	-51	2230	100	760	219	270	220	271	221	273	222	274
20,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
22,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
24,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
26,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
28,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
29,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA – 20 ℃

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	0	-5	2230	100	934	238	231	239	232	240	232	241	233
2000	-4	-9	2230	100	908	236	235	237	236	238	237	239	238
4000	-8	-13	2230	100	880	233	239	234	240	235	241	236	242
6000	-11	-17	2230	100	858	231	244	232	245	233	246	234	247
8000	-15	-21	2230	100	838	229	248	230	250	231	251	232	252
10,000	-19	-25	2230	100	818	226	253	228	254	229	225	230	257
12,000	-23	-29	2230	100	796	224	258	225	259	226	260	227	262
14,000	-26	-33	2230	100	782	222	263	223	264	224	266	225	267
16,000	-30	-37	2230	100	772	220	268	221	270	222	271	223	272
18,000	-34	-41	2230	100	764	216	273	218	275	219	276	220	278
20,000	-37	-45	2230	100	758	213	278	215	280	216	281	218	283
22,000	-41	-49	2230	100	752	210	283	212	285	214	287	215	288
24,000	-45	-53	2096	94	704	202	280	204	283	206	286	208	288
26,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
28,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
29,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA – 10 ºC

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	10	5	2230	100	940	236	233	237	234	238	235	239	236
2000	6	1	2230	100	914	234	238	235	239	236	240	237	240
4000	3	-3	2230	100	888	232	242	233	243	234	244	235	245
6000	-1	-7	2230	100	862	229	247	231	248	232	249	232	250
8000	-5	-11	2230	100	840	227	251	228	253	229	254	230	255
10,000	-9	-15	2230	100	818	225	256	226	257	227	259	228	260
12,000	-12	-19	2230	100	800	222	261	223	262	225	264	226	265
14,000	-16	-23	2230	100	786	220	266	221	268	222	269	223	270
16,000	-20	-27	2230	100	776	217	271	218	273	220	274	221	276
18,000	-24	-31	2230	100	768	214	276	216	278	217	279	218	281
20,000	-27	-35	2230	100	762	211	281	213	283	214	285	215	286
22,000	-31	-39	2119	95	724	204	280	206	283	208	286	210	288
24,000	-35	-43	2007	90	680	196	279	199	282	201	285	202	287
26,000	-40	-47	1851	83	630	187	275	190	279	192	283	195	286
28,000	-44	-51	1717	77	580	177	270	181	275	184	279	186	283
29,000	-46	-52	1628	73	556	172	266	175	272	179	277	181	281
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	20	15	2230	100	946	235	236	236	237	237	238	238	239
2000	16	11	2230	100	920	232	240	234	241	235	242	235	243
4000	13	7	2230	100	892	230	245	231	246	232	247	233	248
6000	9	3	2230	100	866	228	249	229	251	230	252	231	253
8000	5	-1	2230	100	844	225	254	226	255	228	257	228	258
10,000	1	-5	2230	100	822	223	259	224	260	225	262	226	263
12,000	-2	-9	2230	100	800	220	264	222	266	223	267	224	286
14,000	-6	-13	2230	100	786	218	269	219	271	221	272	222	274
16,000	-10	-17	2230	100	776	215	274	216	276	218	277	219	279
18,000	-13	-21	2230	100	772	212	278	213	281	215	282	217	284
20,000	-17	-25	2141	96	734	205	279	207	281	209	283	210	285
22,000	-21	-29	2029	91	698	199	274	200	281	202	284	204	286
24,000	-25	-33	1896	85	652	191	276	193	280	195	283	197	286
26,000	-29	-37	1762	79	604	181	272	184	277	187	280	189	284
28,000	-34	-41	1628	73	558	171	266	175	272	178	277	181	281
29,000	-36	-42	1561	70	536	166	263	170	269	173	275	176	279
31,000	-40	-46	1427	64	494	154	253	159	262	164	270	167	275
33,000	-45	-50	1316	59	452	137	235	148	253	154	262	158	270
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA + 10 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	30	25	2230	100	952	233	238	234	239	235	240	236	241
2000	27	21	2230	100	924	231	243	232	244	233	145	234	246
4000	23	17	2230	100	898	229	247	230	249	231	250	232	251
6000	19	13	2230	100	874	226	252	227	253	228	255	229	256
8000	15	9	2230	100	848	224	257	225	258	226	259	227	261
10,000	12	5	2230	100	826	221	262	222	263	224	265	225	266
12,000	8	1	2230	100	804	219	267	220	269	221	270	222	271
14,000	4	-3	2230	100	790	216	272	217	273	219	275	220	276
16,000	0	-7	2230	100	778	213	277	214	278	216	280	217	282
18,000	-3	-11	2141	96	740	206	277	209	279	210	281	211	283
20,000	-7	-15	2052	92	706	200	277	202	280	204	282	205	285
22,000	-11	-19	1940	87	672	192	276	195	279	197	282	199	285
24,000	-15	-23	1806	81	628	184	273	187	277	189	281	191	284
26,000	-20	-27	1695	76	580	175	268	178	274	181	278	184	282
28,000	-24	-31	1561	70	536	164	262	169	269	172	274	175	279
29,000	-26	-32	1494	67	514	158	257	163	265	167	271	171	277
31,000	-30	-36	1360	61	474	145	245	153	257	158	265	162	272
33,000	-35	-40	1249	56	434	– – –	– – –	140	245	148	258	153	267
35,000	-41	-44	1093	49	388	– – –	– – –	– – –	– – –	134	245	142	259

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA + 20 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	40	35	2230	100	958	323	241	233	242	234	243	235	244
2000	37	31	2230	100	932	229	245	231	247	232	248	233	249
4000	33	27	2230	100	904	229	250	228	251	229	252	230	253
6000	29	23	2230	100	878	225	255	226	256	227	257	228	258
8000	25	19	2230	100	854	222	260	223	261	224	262	225	263
10,000	22	15	2230	100	832	220	265	221	266	222	268	223	269
12,000	18	11	2230	100	810	217	270	218	271	220	273	221	274
14,000	14	7	2185	98	780	212	272	214	274	215	276	217	278
16,000	10	3	2096	94	744	206	273	208	275	210	278	211	279
18,000	6	-1	2029	91	708	200	274	202	276	204	279	206	281
20,000	3	-5	1940	87	678	194	274	196	277	198	280	200	283
22,000	-1	-9	1851	83	646	187	274	190	278	192	281	995	284
24,000	-6	-13	1739	78	604	179	271	182	275	185	279	187	283
26,000	-10	-17	1606	72	558	169	265	173	271	176	276	179	280
28,000	-14	-21	1472	66	514	157	257	162	264	167	271	170	276
29,000	-16	-22	1338	60	472	151	251	157	260	162	268	165	273
31,000	-21	-26	1227	55	434	133	230	146	251	152	261	156	269
33,000	-26	-30	1093	49	392	– – –	– – –	128	229	141	252	147	262
35,000	-29	-34	1026	46	368	– – –	– – –	– – –	– – –	– – –	– – –	136	252

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA + 30 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	51	45	2230	100	964	231	243	231	244	232	245	233	246
2000	47	41	2230	100	936	228	248	229	249	230	250	231	251
4000	43	37	2230	100	910	226	252	227	254	228	255	229	256
6000	39	33	2230	100	882	223	257	224	259	225	260	226	261
8000	36	29	2230	100	858	221	262	222	264	223	265	224	266
10,000	32	25	2208	99	830	216	266	219	268	220	270	221	271
12,000	28	21	2096	94	774	210	266	212	268	213	270	215	271
14,000	24	17	2029	91	740	205	267	206	269	208	272	210	273
16,000	20	13	1962	88	706	199	268	201	271	203	273	204	275
18,000	16	9	1896	85	674	193	269	195	272	197	275	199	277
20,000	12	5	1851	83	650	188	271	190	274	193	277	194	288
22,000	8	1	1762	79	620	181	271	184	275	187	278	189	281
24,000	4	-3	1650	74	580	173	267	176	272	179	276	182	280
26,000	0	-7	1539	69	536	163	261	167	267	171	273	174	278
28,000	-4	-11	1405	63	494	151	252	157	261	162	268	165	274
29,000	-7	-12	1360	61	474	144	244	151	257	157	265	161	272
31,000	-11	-16	1227	55	434	– – –	– – –	138	243	146	257	151	266
33,000	-15	-20	1137	51	402	– – –	– – –	– – –	– – –	133	243	141	258
35,000	-19	-24	1026	46	370	– – –	– – –	– – –	– – –	– – –	– – –	128	243

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1700 RPM

ISA + 37 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	58	52	2230	100	968	229	245	231	246	232	247	232	248
2000	54	48	2230	100	940	227	249	228	251	229	252	230	253
4000	50	44	2230	100	912	225	254	226	255	227	257	228	258
6000	46	40	2230	100	886	222	259	223	260	224	262	225	263
8000	43	36	2185	98	846	217	261	219	263	220	265	221	266
10,000	39	32	2096	94	798	211	262	213	264	215	266	216	268
12,000	35	28	1985	89	744	204	262	206	264	208	266	209	268
14,000	31	24	1918	86	712	200	264	201	266	203	269	205	270
16,000	27	20	1873	84	680	194	265	196	268	198	270	200	273
18,000	23	16	1806	81	650	188	266	191	269	193	271	195	275
20,000	19	12	1762	79	628	183	267	186	271	188	274	190	277
22,000	15	8	1695	76	602	177	267	180	272	183	276	185	279
24,000	11	4	1583	71	562	168	263	172	269	175	274	178	278
26,000	7	0	1472	66	520	158	257	163	264	167	270	170	276
28,000	3	-4	1360	61	478	146	246	153	257	157	265	162	272
29,000	0	-5	1293	58	458	136	234	147	252	152	261	157	2669
31,000	-5	-9	1182	53	420	– – –	– – –	134	234	142	253	148	263
33,000	-9	-13	1093	49	390	– – –	– – –	– – –	– – –	127	236	137	254
35,000	-13	-17	1004	45	358	– – –	– – –	– – –	– – –	– – –	– – –	123	237

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1800 RPM

ISA - 30 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	-10	-15	2185	98	962	244	232	244	232	244	232	244	232
2000	-14	-19	2230	100	946	243	237	244	238	244	238	244	238
4000	-18	-23	2230	100	920	241	242	241	243	242	243	243	244
6000	-21	-27	2230	100	896	238	246	239	247	240	248	241	249
8000	-25	-31	2230	100	874	236	251	237	252	238	253	239	254
10,000	-29	-35	2230	100	854	233	255	235	257	236	258	236	259
12,000	-32	-39	2230	100	834	231	260	232	262	233	263	234	264
14,000	-36	-43	2230	100	820	229	266	230	267	231	268	232	269
16,000	-40	-47	2230	100	808	226	271	228	272	229	273	230	275
18,000	-44	-51	2230	100	798	224	276	225	278	226	279	228	280
20,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
22,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
24,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
26,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
28,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
29,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
31,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
33,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
35,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1800 RPM

ISA - 20 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	0	-5	2230	100	972	243	236	244	236	244	236	244	236
2000	-4	-9	2230	100	948	241	240	242	241	243	242	244	243
4000	-7	-13	2230	100	922	239	245	240	246	241	246	241	247
6000	-11	-17	2230	100	898	236	249	237	250	238	251	239	252
8000	-15	-21	2230	100	876	234	254	235	255	236	256	237	257
10,000	-19	-25	2230	100	856	232	259	233	260	234	261	235	262
12,000	-22	-29	2230	100	834	229	264	230	265	231	266	232	267
14,000	-26	-33	2230	100	822	227	269	228	270	230	272	230	273
16,000	-30	-37	2230	100	810	224	274	226	276	227	277	228	278
18,000	-33	-41	2230	100	800	222	280	223	281	225	283	226	284
20,000	-37	-45	2141	96	764	216	281	217	283	219	285	220	286
22,000	-41	-49	2052	92	726	209	281	211	283	213	285	214	287
24,000	-45	-53	1918	86	680	201	279	203	282	205	285	207	287
26,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
28,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
29,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1800 RPM

ISA - 10 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	10	5	2230	100	978	242	238	243	239	243	240	244	241
2000	7	1	2230	100	950	239	243	240	244	241	245	242	246
4000	3	-3	2230	100	926	237	247	238	248	239	249	240	250
6000	-1	-7	2230	100	900	234	252	236	253	237	254	237	255
8000	-5	-11	2230	100	878	232	257	233	258	234	259	235	260
10,000	-8	-15	2230	100	856	230	262	231	263	232	264	233	265
12,000	-12	-19	2230	100	838	227	267	229	268	230	270	231	271
14,000	-16	-23	2230	100	824	225	272	226	274	227	275	228	276
16,000	-19	-27	2230	100	812	222	278	224	279	225	281	226	282
18,000	-23	-31	2141	96	776	216	279	218	281	219	283	221	284
20,000	-27	-35	2052	92	734	210	279	211	281	213	283	214	285
22,000	-31	-39	1940	87	698	203	278	205	282	207	284	208	286
24,000	-35	-43	1829	82	654	195	277	197	281	200	283	201	286
26,000	-39	-47	1695	76	608	186	274	189	278	191	281	194	284
28,000	-44	-51	1561	70	560	176	269	180	274	183	278	185	282
29,000	-46	-52	1494	67	536	171	265	175	271	178	276	181	280
31,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
33,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
35,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1800 RPM

ISA

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	20	15	2230	100	984	240	241	241	242	242	243	243	244
2000	17	11	2230	100	956	238	246	239	247	240	248	240	248
4000	13	7	2230	100	928	235	250	236	251	237	252	238	253
6000	9	3	2230	100	902	233	255	234	256	235	257	236	258
8000	5	-1	2230	100	880	230	260	232	261	233	262	234	263
10,000	2	-5	2230	100	858	228	265	229	266	230	267	231	269
12,000	-2	-9	2230	100	838	226	270	227	272	228	273	229	274
14,000	-6	-13	2230	100	882	223	275	224	277	226	278	227	280
16,000	-9	-17	2163	97	792	218	278	219	279	221	281	222	283
18,000	-13	-21	2029	91	744	210	276	212	279	213	281	215	282
20,000	-17	-25	1940	87	708	204	277	206	279	207	282	209	284
22,000	-21	-29	1851	83	672	197	277	199	280	201	283	203	285
24,000	-25	-33	1739	78	630	189	275	192	278	194	282	196	284
26,000	-29	-37	1628	73	582	180	271	183	275	186	279	188	282
28,000	-34	-41	1494	67	536	170	265	174	271	177	275	180	279
29,000	-36	-42	1427	64	514	164	260	168	267	172	273	175	277
31,000	-40	-46	1316	59	474	151	248	158	260	163	268	166	273
33,000	-45	-50	1182	53	434	131	225	145	248	152	260	157	268
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1800 RPM

ISA + 10 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	31	25	2230	100	990	238	244	239	245	240	246	241	246
2000	27	21	2230	100	962	236	248	237	249	238	250	239	251
4000	23	17	2230	100	936	234	253	235	254	236	255	237	256
6000	19	13	2230	100	910	231	258	232	259	233	260	234	261
8000	16	9	2230	100	886	229	263	230	264	231	265	232	266
10,000	12	5	2230	100	862	226	268	228	269	229	270	230	272
12,000	8	1	2230	100	840	224	273	225	275	226	276	227	277
14,000	4	-3	2163	97	802	218	275	220	277	221	278	222	280
16,000	0	-7	2052	92	756	212	275	213	277	215	279	216	281
18,000	-4	-11	1962	88	714	205	275	207	277	208	280	210	281
20,000	-7	-15	1873	84	682	199	275	201	278	203	281	204	283
22,000	-11	-19	1784	80	648	191	275	194	278	196	281	198	283
24,000	-15	-23	1673	75	606	183	272	186	276	188	279	191	282
26,000	-20	-27	1539	69	560	174	267	177	272	180	277	183	280
28,000	-24	-31	1405	63	514	163	260	167	267	171	272	174	277
29,000	-26	-32	1360	61	494	149	247	162	263	166	270	170	275
31,000	-31	-36	1227	55	454	133	229	150	253	156	263	161	270
33,000	-36	-40	1115	50	414	– – –	– – –	135	237	145	254	151	264
35,000	-40	-44	1026	46	380	– – –	– – –	– – –	– – –	130	237	140	254

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1800 RPM

ISA + 20 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	41	35	2230	100	994	237	246	238	247	239	248	240	249
2000	37	31	2230	100	968	235	251	236	252	237	253	237	254
4000	33	27	2230	100	940	232	255	233	257	234	258	235	259
6000	29	23	2230	100	914	230	261	231	262	232	263	233	264
8000	26	19	2230	100	890	227	266	228	267	230	268	230	269
10,000	22	15	2208	99	862	224	270	225	271	227	273	228	274
12,000	18	11	2096	94	806	217	270	219	271	220	273	221	275
14,000	14	7	2007	90	762	211	270	213	273	214	274	215	276
16,000	10	3	1918	86	722	205	271	207	273	208	276	210	277
18,000	6	-1	1851	83	682	198	271	200	274	202	276	204	279
20,000	2	-5	1784	80	656	193	273	195	276	197	279	199	281
22,000	-1	-9	1717	77	626	187	273	189	277	191	280	193	283
24,000	-6	-13	1606	72	584	178	270	181	274	184	278	186	282
26,000	-10	-17	1472	66	538	168	264	172	270	175	274	178	279
28,000	-14	-21	1338	60	494	155	253	161	263	165	269	169	274
29,000	-17	-22	1271	57	472	147	245	155	257	160	266	164	272
31,000	-21	-26	1160	52	434	– – –	– – –	142	245	150	258	155	266
33,000	-25	-30	1026	46	380	– – –	– – –	– – –	– – –	137	245	145	259
35,000	-29	-34	982	44	368	– – –	– – –	– – –	– – –	– – –	– – –	132	246

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NORMAL CRUISE POWER

1800 RPM

ISA + 30 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	51	45	2230	100	1000	235	249	237	250	237	251	238	251
2000	47	41	2230	100	972	233	253	234	254	235	255	236	256
4000	43	37	2230	100	944	231	258	232	259	233	261	234	262
6000	40	33	2230	100	916	228	263	229	264	230	265	231	267
8000	36	29	2141	96	868	222	264	224	266	225	267	226	268
10,000	32	25	2052	92	818	216	265	218	267	219	269	220	270
12,000	28	21	1962	88	766	209	265	211	267	213	269	214	271
14,000	24	17	1873	84	724	203	265	205	268	207	270	208	272
16,000	20	13	1806	81	686	197	266	199	269	201	271	203	273
18,000	16	9	1717	77	652	191	267	194	270	196	273	197	275
20,000	12	5	1673	75	626	186	269	189	272	191	275	193	278
22,000	8	1	1628	73	602	181	270	184	274	186	277	188	281
24,000	4	-3	1516	68	560	172	266	175	271	178	275	181	279
26,000	0	-7	1405	63	516	161	259	166	266	170	272	173	276
28,000	-4	-11	1293	58	474	148	247	156	259	160	266	164	272
29,000	-7	-12	1227	55	454	140	237	149	252	155	263	159	270
31,000	-12	-16	1115	50	414	– – –	– – –	134	236	144	253	150	263
33,000	-16	-20	1004	45	384	– – –	– – –	– – –	– – –	128	235	139	253
35,000	-20	-24	914	41	352	– – –	– – –	– – –	– – –	– – –	– – –	123	234

NOT INTENTED FOR REAL FLIGHTS.

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NORMAL CRUISE POWER

1800 RPM

ISA + 37 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	58	52	2230	100	1002	234	250	236	251	236	252	237	253
2000	54	48	2208	99	968	231	254	232	255	233	256	236	257
4000	50	44	2163	97	930	227	257	228	258	229	260	231	261
6000	46	40	2096	94	886	222	259	224	261	225	262	226	263
8000	43	36	2029	91	838	217	260	218	262	220	264	221	265
10,000	39	32	1962	88	792	211	262	213	264	214	266	215	267
12,000	35	28	1851	83	738	204	261	206	264	208	266	209	268
14,000	31	24	1784	80	700	198	262	200	265	202	267	203	269
16,000	27	20	1717	77	664	192	262	194	265	196	268	198	270
18,000	23	16	1650	74	630	186	263	189	266	191	269	193	272
20,000	19	12	1606	72	606	181	265	184	269	186	272	188	275
22,000	15	8	1561	70	584	176	266	179	271	182	274	184	278
24,000	11	4	1450	65	544	167	262	171	268	174	273	177	277
26,000	7	0	1360	61	502	156	254	162	263	166	269	169	274
28,000	3	-4	1227	55	460	142	240	151	254	156	263	160	270
29,000	1	-5	1182	53	440	130	223	143	247	151	259	156	267
31,000	-5	-9	1048	47	400	– – –	– – –	125	223	139	247	146	260
33,000	-9	-13	981	44	370	– – –	– – –	– – –	– – –	121	225	134	249
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1800 RPM

ISA - 30 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	-10	-15	2185	98	962	244	232	244	232	244	232	244	232
2000	-14	-19	2230	100	946	243	237	244	238	244	238	244	238
4000	-18	-23	2230	100	920	241	242	241	243	242	243	243	244
6000	-22	-27	2230	100	896	238	246	239	247	240	248	241	249
8000	-25	-31	2230	100	874	236	251	237	252	238	253	239	254
10,000	-29	-35	2230	100	854	233	255	235	257	236	258	236	259
12,000	-33	-39	2230	100	834	231	260	232	262	233	263	234	264
14,000	-36	-43	2230	100	820	229	265	230	267	231	268	232	269
16,000	-40	-47	2230	100	808	226	271	228	272	229	273	230	275
18,000	-44	-51	2230	100	798	224	276	226	278	226	279	228	280
20,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
22,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
24,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
26,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
28,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
29,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
31,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
33,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
35,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1800 RPM

ISA - 20 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	0	-5	2230	100	972	243	236	244	236	244	236	244	236
2000	-4	-9	2230	100	948	241	240	242	241	243	242	244	243
4000	-8	-13	2230	100	922	239	245	240	246	241	246	241	247
6000	-11	-17	2230	100	898	236	249	237	250	238	251	239	252
8000	-15	-21	2230	100	876	234	254	235	255	236	256	237	257
10,000	-19	-25	2230	100	856	232	259	233	260	234	261	235	262
12,000	-23	-29	2230	100	834	229	264	230	265	231	266	232	267
14,000	-26	-33	2230	100	822	227	269	228	270	230	272	230	273
16,000	-30	-37	2230	100	810	224	274	226	276	227	277	228	278
18,000	-34	-41	2230	100	800	222	280	223	281	225	283	226	284
20,000	-37	-45	2230	100	792	220	285	221	287	222	289	223	290
22,000	-41	-49	2141	96	726	213	286	214	288	216	290	217	292
24,000	-45	-53	2007	90	704	204	284	206	286	208	289	210	291
26,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
28,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
29,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
31,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
33,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -
35,000	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -	- - -

NOT INTENTED FOR REAL FLIGHTS.

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MAXIMUM CRUISE POWER

1800 RPM

ISA - 10 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	10	5	2230	100	978	242	238	243	239	243	240	244	241
2000	6	1	2230	100	950	239	243	240	244	241	245	242	246
4000	3	-3	2230	100	926	237	247	238	248	239	249	240	250
6000	-1	-7	2230	100	900	234	252	236	253	237	254	237	255
8000	-5	-11	2230	100	878	232	257	233	258	234	259	235	260
10,000	-9	-15	2230	100	856	230	262	231	263	232	264	233	265
12,000	-12	-19	2230	100	838	227	267	229	268	230	270	231	271
14,000	-16	-23	2230	100	824	225	272	226	274	227	275	228	276
16,000	-20	-27	2230	100	812	222	278	224	279	225	281	226	282
18,000	-24	-31	2230	100	802	220	283	221	285	223	287	223	288
20,000	-27	-35	2141	96	764	213	284	215	286	217	288	218	289
22,000	-31	-39	2029	91	726	206	284	208	286	210	288	212	290
24,000	-35	-43	1918	86	682	199	282	201	285	203	288	204	290
26,000	-40	-47	1784	80	632	190	279	192	283	195	286	197	288
28,000	-44	-51	1628	73	582	181	274	183	279	186	283	188	286
29,000	-46	-52	1561	70	556	175	271	178	276	181	280	184	284
31,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
33,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

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MAXIMUM CRUISE POWER

1800 RPM

ISA

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	20	15	2230	100	946	240	241	241	242	242	243	243	244
2000	16	11	2230	100	920	238	246	239	247	240	248	240	248
4000	13	7	2230	100	892	235	250	236	251	237	252	238	253
6000	9	3	2230	100	866	233	255	234	256	235	257	236	258
8000	5	-1	2230	100	844	230	260	232	261	233	262	234	263
10,000	1	-5	2230	100	822	228	265	229	266	230	267	231	269
12,000	-2	-9	2230	100	800	226	270	227	272	228	273	229	274
14,000	-6	-13	2230	100	786	223	275	224	277	226	278	227	280
16,000	-10	-17	2230	100	776	221	281	222	283	223	284	224	285
18,000	-13	-21	2119	95	772	213	281	216	283	217	285	218	287
20,000	-17	-25	2029	91	734	207	282	209	284	211	286	212	288
22,000	-21	-29	1940	87	698	201	282	203	284	204	287	206	289
24,000	-25	-33	1829	82	654	192	280	195	283	197	286	199	288
26,000	-29	-37	1695	76	606	184	276	186	280	189	284	191	287
28,000	-34	-41	1561	70	560	174	271	177	276	180	281	183	284
29,000	-36	-42	1494	67	538	169	268	172	273	176	278	178	283
31,000	-40	-46	1383	62	496	157	258	162	267	167	274	170	279
33,000	-45	-50	1249	56	454	140	240	150	257	157	267	161	274
35,000	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –	– – –

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MAXIMUM CRUISE POWER

1800 RPM

ISA + 10 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	31	25	2230	100	990	238	244	239	245	240	246	241	246
2000	27	21	2230	100	962	236	248	237	249	238	250	239	251
4000	23	17	2230	100	936	234	253	235	254	236	255	237	256
6000	19	13	2230	100	910	231	258	232	259	233	260	234	261
8000	16	9	2230	100	886	229	263	230	264	231	265	232	266
10,000	12	5	2230	100	862	226	268	228	269	229	270	230	272
12,000	8	1	2230	100	840	224	273	225	275	226	276	227	277
14,000	5	-3	2230	100	824	221	278	223	280	224	282	225	283
16,000	1	-7	2141	96	782	215	279	216	281	218	283	219	284
18,000	-3	-11	2029	91	742	208	279	210	282	212	284	213	286
20,000	-7	-15	1962	88	708	202	280	204	283	206	285	207	287
22,000	-11	-19	1851	83	672	195	279	197	282	199	285	201	288
24,000	-15	-23	1717	77	628	187	277	189	281	192	284	194	287
26,000	-19	-27	1606	72	582	177	273	181	277	183	281	186	285
28,000	-24	-31	1494	67	536	167	267	171	273	175	278	177	282
29,000	-26	-32	1427	64	516	161	262	166	270	170	275	173	280
31,000	-30	-36	1293	58	474	148	250	156	262	161	270	164	276
33,000	-35	-40	1182	53	436	– – –	– – –	143	250	150	263	155	271
35,000	-39	-44	1093	49	402	– – –	– – –	– – –	– – –	137	249	145	263

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MAXIMUM CRUISE POWER

1800 RPM

ISA + 20 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	41	35	2230	100	994	337	246	238	247	239	248	240	249
2000	37	31	2230	100	968	235	251	236	252	237	253	237	254
4000	33	27	2230	100	940	232	255	233	257	234	258	235	259
6000	29	23	2230	100	914	230	261	231	262	232	263	233	264
8000	26	19	2230	100	890	227	266	228	267	230	268	230	269
10,000	22	15	2230	100	868	225	271	226	272	227	273	228	275
12,000	18	11	2141	96	820	219	272	220	274	222	275	223	277
14,000	14	7	2074	93	780	213	274	215	276	216	277	218	279
16,000	10	3	2007	90	744	208	275	210	277	211	279	213	281
18,000	7	-1	1918	86	710	202	276	204	279	206	281	207	283
20,000	3	-5	1851	83	678	196	277	198	280	200	283	202	285
22,000	-1	-9	1784	80	648	190	278	192	281	194	284	996	287
24,000	-5	-13	1673	75	606	181	275	184	279	187	283	189	286
26,000	-9	-17	1539	69	560	172	270	175	275	178	279	181	283
28,000	-14	-21	1405	63	516	161	262	165	269	169	275	172	279
29,000	-16	-22	1338	60	494	154	255	160	265	164	272	168	277
31,000	-21	-26	1227	55	454	137	237	148	255	154	266	1559	273
33,000	-26	-30	1115	50	416	– – –	– – –	132	237	143	256	150	267
35,000	-30	-34	1026	46	382	– – –	– – –	– – –	– – –	127	236	138	256

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MAXIMUM CRUISE POWER

1800 RPM

ISA + 30 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	51	45	2230	100	1000	235	248	236	250	237	251	238	251
2000	47	41	2230	100	972	233	253	234	254	235	255	236	256
4000	43	37	2230	100	944	231	258	232	259	233	261	234	262
6000	40	33	2230	100	918	228	263	229	264	230	266	231	267
8000	36	29	2185	98	878	224	266	225	267	226	269	227	270
10,000	32	25	2096	94	830	218	267	219	269	221	270	222	272
12,000	28	21	1985	89	774	211	267	212	269	214	270	215	272
14,000	24	17	1918	86	740	205	268	207	271	209	273	210	275
16,000	20	13	1873	84	706	200	270	202	273	204	275	205	277
18,000	16	9	1806	81	674	195	272	197	275	199	277	201	279
20,000	13	5	1739	78	650	190	274	182	277	194	280	196	283
22,000	9	1	1695	76	622	184	274	187	278	189	281	191	284
24,000	5	-3	1583	71	580	175	271	179	276	181	280	184	283
26,000	0	-7	1472	66	538	166	266	170	272	173	277	176	281
28,000	-4	-11	1360	61	496	154	257	161	266	164	273	168	278
29,000	-6	-12	1293	58	476	147	248	154	261	159	269	163	276
31,000	-11	-16	1182	53	436	– – –	– – –	141	248	149	262	154	270
33,000	-15	-20	1070	48	404	– – –	– – –	– – –	– – –	135	248	144	262
35,000	-19	-24	981	44	370	– – –	– – –	– – –	– – –	– – –	– – –	130	247

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MAXIMUM CRUISE POWER

1800 RPM

ISA + 37 °C

NOTES: IOAT, TORQUE, AND FUEL FLOW BASED ON 12,000 POUNDS.
TO OBTAIN FUEL FLOW PER ENGINE, DIVIDE THE TOTAL FUEL FLOW VALUE FOR THE CORRESPONDING ALTITUDE BY TWO.

PRESSURE ALT	IOAT	OAT	TORQUE PER ENGINE		TOTAL FUEL FLOW	AIRSPEED ~ KNOTS							
						13,000 LB		12,000 LB		11,000 LB		10,000 LB	
FEET	C	C	FT-LB	%	LB/HR	IAS	TAS	IAS	TAS	IAS	TAS	IAS	TAS
SL	58	52	2230	100	1002	234	250	236	251	236	252	237	253
2000	54	48	2230	100	974	232	255	233	256	234	257	235	258
4000	50	44	2208	99	940	228	258	230	260	231	261	232	262
6000	46	40	2141	96	894	223	261	225	262	226	264	227	265
8000	43	36	2052	92	846	218	262	219	264	221	265	222	267
10,000	39	32	1985	89	798	212	263	214	265	215	267	216	268
12,000	35	28	1873	84	746	205	263	207	265	209	267	210	269
14,000	31	24	1829	82	712	200	264	202	267	204	269	205	271
16,000	27	20	1762	79	682	195	266	197	269	199	272	201	274
18,000	23	16	1717	77	650	190	268	192	271	194	274	196	276
20,000	19	12	1673	75	628	185	2670	187	274	190	277	192	280
22,000	15	8	1628	73	602	179	271	182	275	185	279	187	282
24,000	11	4	1516	68	564	171	268	174	273	177	277	180	281
26,000	7	0	1405	63	522	161	262	166	269	169	274	172	279
28,000	3	-4	1293	58	480	148	250	156	262	160	269	164	275
29,000	0	-5	1249	56	460	139	239	149	256	155	266	159	273
31,000	-4	-9	1115	50	422	– – –	– – –	134	239	144	257	150	267
33,000	-8	-13	1048	47	390	– – –	– – –	– – –	– – –	130	241	140	258
35,000	-12	-17	959	43	360	– – –	– – –	– – –	– – –	– – –	– – –	125	242

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OPERATING CONDITION	SHP	TORQUE % (1)	MAXIMUM OBSERVED ITT °C	GAS GENERATOR RPM N_1		PROP RPM N_2	OIL PRESS PSI (2)	OIL TEMP °C (3) (4)
				RPM	%			
STARTING	– – –	– – –	1000 (5)	– – –	– – –	– – –	– – –	-40 (min)
LOW IDLE	– – –	– – –	750 (6)	22,875	61(min)	1180	60 (min)	-40 to 99
HIGH IDLE	– – –	– – –	– – –	– – –	(7)	– – –	– – –	-40 to 99
TAKEOFF and MAX CONT	850	100	800	38,100	101.5	2000	100 to 135	0 to 99
MAX CRUISE	850	100 (8)	800	38,100	101.5	2000	100 to 135	0 to 99
CRUISE CLIMB and REC (NORMAL) CRUISE	850	100 (8)	770	38,100	101.5	2000	100 to 135	0 to 99
MAX REVERSE (9)	800	– – –	750	– – –	88	1900	100 to 135	0 to 99
TRANSIENT	– – –	123 (5)	850	38,500 (10)	102.6 (10)	2200 (5)	200	0 to 104 (11)

TABLE NOTES:

(1) Torque limit applies within range of 1600 - 2000 propeller RPM (N2). Below 1600 propeller RPM, torque is limited to 49%.

(2) When gas generator speeds are above 27,000 RPM (72% N1) and oil temperatures are between 60 °C and 71°C, normal oil pressures are: 100 to 135 psi below 21,000 feet; and 85 to 135 psi at 21,000 feet and above.
During extremely cold starts, oil pressure may reach 200 psi. Oil pressure between 60 and 85 psi is undesirable. It should be tolerated only for the completion of the flight, and then only at a reduced power setting not exceeding 49% torque. Oil pressure below 60 psi is unsafe. It requires that either the engine be shut down, or that a landing be made at the nearest suitable airport, using the minimum power required to sustain flight. Fluctuations of ± 10 psi are acceptable.

(3) A minimum oil temperature of 55 °C is recommended for fuel heater operation at takeoff power.

(4) Oil temperature limits are -40 °C and 99 °C. However, temperatures of up to 104 °C are permitted for a maximum time of 10 minutes.

(5) These values are time limited to 5 seconds.

(6) High ITT at ground idle may be corrected by reducing accessory load or increasing N1 RPM.

(7) At approximately 70% N1.

(8) Cruise torque values vary with altitude and temperature.

(9) This operation is time limited to 1 minute.

(10) These values are time limited to 10 seconds.

(11) Values above 99 °C are time limited to 5 minutes.

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